

**Amendments to the Claims:**

This listing of claims will replace all prior versions and listing of claims in the application.

**Listing of Claims:**

1-11. (canceled).

12. (currently amended): A gas turbine combustor comprising:

a plurality of fuel nozzles;

an air hole forming body having a plurality of air holes formed therein;

and a combustion chamber,

wherein said fuel nozzles and said air holes are formed so that fuel and air are jetted from said fuel nozzles and air holes into said combustion chamber as a plurality of coaxial jet flows, and

a part of all of said fuel nozzles are formed in double construction so that spraying of liquid fuel and jetting of gaseous fuel can be ~~switched or~~ combined.

13. (currently amended): A gas turbine combustor comprising:

a plurality of fuel nozzles;

an air hole forming body having a plurality of air holes formed therein;

and a combustion chamber,

wherein said fuel nozzles and said air holes are formed so that fuel and air are jetted from said fuel nozzles and said air holes into said combustion chamber as a plurality of coaxial jet flows,

fuel jetting holes of said fuel nozzles are arranged coaxially with or at positions close to said air holes, respectively, so that fuel jet flows are jetted from said fuel jet holes to around inlet centers of said air holes and the fuel jet flows and air circulation flows enclosing the fuel jet flows are jetted from outlets of said air holes into said combustion chamber as the plurality of coaxial flows, and

a part of all of said fuel nozzles are formed in double construction so that spraying of liquid fuel and jetting of gaseous fuel can be switched or combined.

14. (previously presented): A gas turbine combustor according to claim 12, wherein said plurality of fuel nozzles are partitioned in a plurality of fuel supply systems, and a control system is provided for controlling independently a fuel flow rate in each of said plurality of fuel supply systems according to a load on the gas turbine.

15. (previously presented): A gas turbine combustor comprising:  
a plurality of fuel nozzles;  
an air hole forming body having a plurality of air holes formed therein;  
and a combustion chamber,

wherein said fuel nozzles and said air holes are formed so that fuel and air are jetted from said fuel nozzles and said air holes into said combustion chamber as a plurality of coaxial jet flows, and

a part or all of said fuel nozzles are formed in double construction, an inside of said double construction being constructed so as to be supplied with liquid fuel and an outside to be supplied with gaseous fuel.

16.-19. (withdrawn).